an actuator entirely disposed in an area between said primary flow duct and said secondary flow duct, said actuator attached to said door and configured to slide said door.

2. The nozzle assembly of claim **1**, further comprising:

said actuator connected to the door and operative to position the door such that the effective size of the nozzle exit area is established.

3. The nozzle assembly of claim **2**, wherein the actuator is operative to translate the door in both fore and aft directions parallel to a longitudinal axis of the gas turbine engine.

4. The nozzle assembly of claim **2**, wherein the nozzle is a third stream exhaust nozzle.

5. The nozzle assembly of claim **1**, further comprising:

a rail engaging the door and operative to provide alignment of the door during positioning.

6. The nozzle assembly of claim **1**, further comprising:

at least one stiffening rib contacting the door and operative to provide structural support to the door.

7. The nozzle assembly of claim **1**, wherein said area is radially inward of said secondary flow duct and radially outward of said primary flow duct.

8. The nozzle assembly of claim **1**, wherein said actuator is entirely disposed upstream of said door.

9. The nozzle assembly of claim **1**, wherein said actuator is configured to pull the door upstream in a fore direction, and wherein said actuator is configured to push the door downstream in an aft direction.

10. A gas turbine engine comprising:

an engine duct structure and an inner structure which at least partially define a secondary flow path for a secondary flow and a primary flow path for a primary flow, said secondary flow path defined at least partially around a perimeter of said primary flow path;

a secondary flow duct with a two dimensional secondary nozzle to communicate said secondary flow therethrough;

a primary flow duct with a two dimensional primary nozzle to communicate said primary flow therethrough, said two dimensional primary nozzle adjacent to said two dimensional secondary nozzle; and

a door axially slidable relative to a passage in communication with said secondary flow path to regulate said secondary flow though said passage; and

an actuator entirely disposed in an area between said primary flow duct and said secondary flow duct, said actuator attached to said door and configured to slide said door.

11. The gas turbine engine of claim **10**, further comprising said actuator connected to the door and operative to position the door.

12. The gas turbine engine of claim **11**, wherein the actuator is operative to translate the door in both fore and aft directions.

13. The gas turbine engine of claim **10**, wherein the nozzle assembly is a third stream exhaust nozzle assembly.

14. The gas turbine engine of claim **10**, further comprising at least one stiffening rib operative to provide structural support to the door.

15. The gas turbine engine of claim **14**, wherein the at least one stiffening rib is located on a non-gas path side of the door.

16. The gas turbine engine of claim **10**, further comprising a rail operative to provide alignment of the door.

17. The gas turbine engine of claim **10**, wherein the engine is a turbofan gas turbine engine.

18. The gas turbine engine of claim **10**, wherein the door is operative to not affect the gas path with respect to yaw.

19. The gas turbine engine of claim **10**, further comprising a plenum located on a non-gas path side of the door, the plenum being operative to pressurize in order to reduce an actuation load of the door.

20. The gas turbine engine of claim **10**, wherein said primary flow includes at least a combustion core gas exhaust flow sourced from a turbine section of the gas turbine engine.

21. The gas turbine engine of claim **10**, wherein said secondary flow is different than said combustion core gas exhaust flow.

22. The gas turbine engine of claim **10**, wherein said two dimensional secondary nozzle is downstream of two dimensional primary nozzle.

23. The gas turbine engine of claim **10**, wherein said two dimensional secondary nozzle is adjacent to said two dimensional primary nozzle.

24. The gas turbine engine of claim **10**, wherein said area is radially inward of said secondary flow duct and radially outward of said primary flow duct.

25. The gas turbine engine of claim **10**, wherein said actuator is entirely disposed upstream of said door.

26. The gas turbine engine of claim **10**, wherein said actuator is configured to pull the door upstream in a fore direction, and wherein said actuator is configured to push the door downstream in an aft direction.

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